

Given: $\gamma(\theta) = V_\infty [1.0 \sin\theta + 0.4 \sin 2\theta]$

General case: $\gamma(\theta) = 2V_\infty \left[A_0 \frac{1+\cos\theta}{\sin\theta} + A_1 \sin\theta + A_2 \sin 2\theta + \dots \right]$

$\therefore A_0 = 0, A_1 = 0.5, A_2 = 0.2, A_3 = A_4 = \dots = 0$

a) $C_L = \pi [2A_0 + A_1] = 0.5\pi = 1.78$

$C_{m/c4} = \frac{\pi}{4} [A_2 - A_1] = -0.3 \frac{\pi}{4} = -0.2356$

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Alternative: $\Delta p = \rho V_\infty \gamma = \rho V_\infty^2 [1.0 \sin\theta + 0.4 \sin 2\theta]$
 (the long way)

$L' = \int_0^c \Delta p dx = \int_0^\pi \Delta p \frac{c}{2} \sin\theta d\theta = \int_0^\pi \rho V_\infty^2 \frac{c}{2} [1.0 \sin\theta + 0.4 \sin 2\theta] \sin\theta d\theta$

$L' = \frac{1}{2} \rho V_\infty^2 c \cdot 1.0 \cdot \frac{\pi}{2} \rightarrow C_L = \frac{L'}{\frac{1}{2} \rho V_\infty^2 c} = \frac{\pi}{2} = 1.78$

$M'_{c/4} = \int_0^c \Delta p \cdot \left(\frac{c}{4} - x\right) dx = \int_0^\pi \Delta p \left(\frac{c}{4} - \frac{c}{2}(1 - \cos\theta)\right) \frac{c}{2} \sin\theta d\theta$

$= \int_0^\pi \rho V_\infty^2 \frac{c^2}{2} [1.0 \sin\theta + 0.4 \sin 2\theta] \left(-\frac{1}{4} + \frac{1}{2} \cos\theta\right) \sin\theta d\theta$

$= \int_0^\pi \rho V_\infty^2 \frac{c^2}{2} [1.0 \sin\theta + 0.4 \sin 2\theta] \left[-\frac{1}{4} \sin\theta + \frac{1}{4} \sin 2\theta\right] d\theta$

$M'_{c/4} = \frac{1}{2} \rho V_\infty^2 c^2 \left[-0.25 \frac{\pi}{2} + 0.1 \frac{\pi}{2}\right] = \frac{1}{2} \rho V_\infty^2 c^2 \left(-0.15 \frac{\pi}{2}\right) \rightarrow C_{m/c4} = \frac{M'_{c/4}}{\frac{1}{2} \rho V_\infty^2 c^2} = -0.2356$

b) A_n 's are also related to airfoil slope:

$x - \frac{dz}{dx} = A_0 - \sum A_n \cos n\theta$ for this case

$\frac{dz}{dx} = x - A_0 + \sum A_n \cos n\theta = x + 0.5 \cos\theta + 0.2 \cos 2\theta$

At $\frac{x}{c} = \frac{1}{2}$ (midchord), $\theta = \frac{\pi}{2}$, $\cos\theta = 0$, $\cos 2\theta = -1$

$\therefore \frac{dz}{dx} = x - 0.2 = 0.1 - 0.2 = -0.1$

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